

A MODEL FOR FATIGUE CRACK GROWTH IN THE PARIS REGIME UNDER THE VARIABILITY OF CYCLIC HARDENING AND ELASTIC PROPERTIES

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Abstract

Over the last 60 years, several models have been developed governing different zones of fatigue crack growth from the threshold zone to final failure. The best known model is the Paris law and a number of its based on mechanical, metallurgical and loading parameters governing the propagation of cracks. This paper presents an analytical model developed to predict the fatigue crack propagation rate in the Paris regime, for different material properties, yield strength (σ_y), Young's modulus (E) and cyclic hardening parameters (K' , n') and their influence by variability. The cyclic plastic deformation at a crack tip or any other cyclic hardening rule may be used to reach this objective, for to investigate this influence, these properties of the model are calibrated using available experimental data in the literature. This FCGR model was validated on Al-alloys specimens under constant amplitude load and shows good agreement with the experimental results.

Keywords: cyclic plastic strain, elastic properties, cyclic hardening parameters, Young's modulus, variability, crack tip, fatigue crack growth, constants of the Paris law, Al-alloys.

1. INTRODUCTION

The fatigue crack growth resistance of a material depends upon a number of factors, such as its composition, mechanical properties and heat treatment conditions, external loading and the environment. Several theoretical models have been proposed to predict the fatigue crack growth (FCG) process using solid mechanics-based theoretical tools and fundamental mechanical properties.

The understanding of the mechanisms governing fatigue crack growth has made significant advances since Paris [1] proposed a law that relates the crack growth rate da / dN and the stress intensity factor range ΔK :

$$\frac{da}{dN} = C(\Delta K)^m \quad (1)$$

Experimentally, the exponent m given in the Paris law can vary between 2 and 6 for most metals and alloys [2]. In Paris law there is no evidence of the mechanical parameters effects. Several authors have attempted to integrate other parameters such as the R -ratio ($R = \sigma_{\min} / \sigma_{\max}$), critical stress intensity factor K_{IC} , etc. Walker [3] reported on the effect of the R -ratio on crack propagation for aluminum alloys, concluding that increasing R resulted in an increased growth rate. For according the model of Walker [3], the influence of the mean stress and R -ratio on fatigue crack propagation in aluminum alloys was suggested by [4]. Paris and Walker equation works well in the Paris region of crack growth, but does not address the asymptotic behavior in the unstable region. To address this behavior, Forman [5] proposed a new relationship that reflects the fracture by incorporating the stress ratio and the critical stress intensity factor K_{IC} given by:

$$\frac{da}{dN} = \frac{C(\Delta K)^m}{(1-R)(K_{IC} - K_{\max})} \quad (2)$$

As K_{\max} approaches K_{IC} , the denominator approaches zero and da/dN tends toward infinity. This equation has been verified many times for aluminum alloys [3], [6]–[8].

Elber [9] was the first researcher to introduce the concept of crack closure in fatigue crack growth. This concept reveals the premature contact of the crack faces during the unloading portion of the loading cycle while some tensile load is still applied. Elber found that the fatigue crack growth rate was a function of a new factor he called effective stress intensity factor range ΔK_{eff} .

In general, the phenomenon of crack closure has been widely accepted as a critical mechanism influencing many aspects of the behavior of fatigue cracks in metallic materials, including variable amplitude loading [10], R -ratio effects, crack size, microstructure [11], cyclic plastic strain [12], thermo-mechanical [13], environment [14], residual stress [15] and the magnitude of the fatigue threshold [16]. The higher fatigue crack growth rates and lower thresholds in the alloy of Magnesium may be attributed to a much smoother fracture surface morphology and lower roughness-induced crack closure and crack deflection [17]. The effects of grain size, also spanning the fatigue crack growth [18].

Besides this mechanically based concept, other approaches based on energy consideration [19]–[21] or on micro mechanisms acting at the crack tip have been developed. In these approaches some authors attempted to express the

crack growth rate by explaining the effects of different parameters by means of theoretical models based on the low cycle fatigue and cyclic strain hardening [19], [22].

Noroozi et al. [23], [24] proposed the FCG equation in terms of a unified two-parameter driving force assuming predominately plastic material deformation at the crack tip while modeling the high fatigue crack growth rate, and the elastic behavior for the near threshold fatigue crack growth. Fatigue crack growth expressions derived for the two different material behaviors [23], [24].

Duggan [25] focused on the deformation of a volume element located at the crack-tip. This volume element is only subjected to elastic deformations and so the effect of softening or of hardening is disregarded. Duggan [25] has assumed that propagation occurs when the Manson-Coffin and Miner laws are satisfied at the same time. He expressed the crack growth rate in terms of the elastic modulus E , toughness K_{IC} , and ductility (σ'_f, ϵ'_f) coefficient and exponent:

$$\frac{da}{dN} = \frac{1}{\epsilon'_f} \left[\frac{2\sqrt{\pi}/32}{\sigma'_f E (K_{IC} - K_{\max})} \left(1 - \frac{\Delta K}{K_{IC}} \right) \right]^{\frac{1}{\epsilon'_f}} \Delta K^{\left(\frac{2}{\epsilon'_f} \right)} \quad (3)$$

Radon [26] proposed a crack growth model for the near threshold region (ΔK_{th}) which incorporates mechanical, cyclic and fatigue properties of the material, and a cyclic plastic strain $(\Delta \epsilon_p)$ based on the failure criterion. He also introduced an effective stress intensity factor range (ΔK_{eff}) which characterizes the crack tip opening displacement and the strains immediately ahead of the crack tip. The cyclic plastic strain range $(\Delta \epsilon_p)$ at the crack front is given by the expression:

$$\Delta \epsilon_p = \frac{2\sigma_y}{E} \left[\frac{\Delta K^2}{4\pi (1+n') \sigma_y^2 a} \right]^{1/1+n'} \quad (4)$$

Where: σ_y , E and n' are the yield strength, Young's modulus and cyclic hardening exponent, respectively.

Radon [26] obtained a simple model of cyclic crack propagation in the threshold region and developed an expression for the crack growth rate, given below:

$$\frac{da}{dN} = \frac{2^{n'-1} (1-2\nu)^2}{\pi (1+n')} \frac{(\Delta K_{eff}^2 - \Delta K_{th}^2)}{\sigma_y^{1-n'} (E \epsilon'_f)^{1+n'}} \quad (5)$$

Lal et al. [27] developed a model that integrates some notions, such as the effective strain intensity factor defined by Elber. These authors calculated the monotonous and cyclic plasticized zones:

$$r_{pm} = \left(\frac{\Delta K}{2\sigma_e} \right)^{1+n'} . a^{1-n'/2} \quad (6)$$

This relationship is obtained from the calculation performed under a simple traction replacing the monotonous work hardening exponent by the cyclic hardening exponent and K by ΔK .

If we apply Elber's hypothesis, Lal et al. [27] presumed that a single part of a cycle was necessary to close the crack, and thus to create the plasticized zone, the other part of the cycle inducing the grounding of the crack lips. Then they used the definition of the monotonous plastic zone that was calculated previously, and by replacing ΔK with ΔK_{eff} they obtained the dimensions of the cyclic plasticized zone:

$$r_{pc} = r_{pm} . U^{1+n'} \quad (7)$$

Using the hypothesis suggested by Tomkins [28], the fatigue crack growth model is described as the function of $\Delta \epsilon_p$ and r_{pc} :

$$\frac{da}{dN} = \Delta \epsilon_p . r_{pc} \quad (8)$$

Where the Tomkins model often used for low cycle fatigue tests under significant plasticity could not explain the results for the variation in crack length nor the variation in loading parameters.

In order to calculate $\Delta \epsilon_p$, they used the equation $\Delta \sigma_N = K' . (\Delta \epsilon_p / 2)^{n'}$ which describes the hysteresis loops, where: $\Delta \sigma_N$ is the variation of stress in the ligament. They expressed the plastic deformation as an amplitude function of the stress intensity factor:

$$\Delta \epsilon_p = \left[\frac{\Delta K}{\sqrt{\pi} . a . K' (1 - a/w)} \right]^{1/n'} \quad (9)$$

The propagation rate is then equal to:

$$\frac{da}{dN} = \frac{(0.5U)^{1+n'} (\sqrt{w})^{1-n'-1/n'}}{(K' \sqrt{\pi})^{1/n'} \sigma_y^{1+n'}} \left(\frac{(\sqrt{a/w})^{n'(1-n')}}{\sqrt{a/w} (1 - a/w)} \right)^{1/n'} \Delta K^{1+n'+1/n'} \quad (10)$$

Two of the parameters of equation (10): (a) and (w), are geometric and easily measurable. The other three U, σ_e, K' and n' depend on the material. The rate decreases when the elasticity limit increases, and it increases if R increases, which matches the experimental results. The parameters σ_e, K' and n' might be easily calculated, but U (except for some aluminum alloys) is less well known. However, there is not enough experimental evidence to determine the variation of the rate as a function of crack length. This equation has been verified for many materials (aluminum, mild steel, stainless steel and copper) and the calculated and measured values of the parameters C and m of Paris' law agree with it.

Models based on the calculation of the crack tip with hardening have been developed by many researchers [19], [26], [27], [29], [30]. These models rely on calculating the plasticized monotonic and cyclical areas created by the cyclical stress; it is generally assumed that the crack propagation is proportional to the energy lost in the plasticized zones. All these models use hysteresis loops to evaluate plastic deformation. The macroscopic behavior characterized by the hysteresis loop is introduced into these models to describe the microscopic behavior at the crack tip.

In a recent study, Shi et al. [30], [31] have developed an analytical fatigue crack growth rate model based on the crack closure expression, effective cyclic plastic zone and the low cycle fatigue properties. Comparative results demonstrate that the fatigue crack growth rate estimated by the theoretical model closely approximates the experimental results. The predicted model is limited to isotropic materials.

Tzamtzis et al. [32] have developed a model to predict the fatigue crack propagation rate under mode I loading in 2024 aluminum alloy. The model considers local cyclic hardening behavior in the heat affected zone (HAZ) to analyze crack growth. The analytical results showed that cyclic hardening at the crack tip can be used successfully with this model to predict FCG in a material at overage condition associated with a location in the Friction Stir Welding heat affected zone (HAZ).

$$\frac{da}{dN} = a^{\frac{2c+3}{2c+2}} \left[\frac{4\sqrt{\pi} E 2^{(c+1/2)} (1-n') \epsilon'_f}{(1-\nu) K_{IC}^2 (1+n')} \right]^{1/(c+1)} \Delta K^{1/(c+1)} \quad (11)$$

where, parameters K_{IC} , c , ϵ'_f , E , can be determined experimentally.

All these models cannot be applied generally; each describes a situation under the influence of certain intrinsic parameters (Young's modulus, grain size, yield strength, toughness) and extrinsic (specimen dimensions, and environmental

effects). They become unsuitable as soon as a parameter of the experience varies. The most of the proposed models of fatigue growth remain essentially empirical, and hence their constants often vary more than expected.

In this paper, we propose an analytical model relating the fatigue crack propagation rate in the Paris regime with different material properties effects at the crack tip, such as cyclic hardening (K' , n') parameters and elastics properties (E , σ_y), and their influence by variability, based on the properties calibrated using available experimental data in the literature. This study introduces the hardening parameter (K' , n') directly in the Paris law parameters (C , m), which the authors consider as an advantage. The details are presented below.

2. ANALYTICAL MODEL

This work focuses on a model of fatigue crack growth, using the assumptions of plastic strain at the fatigue crack tip and the Tomkins model [28]. We have developed a model for fatigue crack growth that varies according to elastic (E , σ_y) and cyclic hardening (K' , n') properties.

The modeling of fatigue crack growth requires good understanding of the evolution of the behavior of a material under cyclic loading. Thus, the cyclic plastic stress/strain response at fatigue crack tips was studied [33], [34], as described in Figure 1.

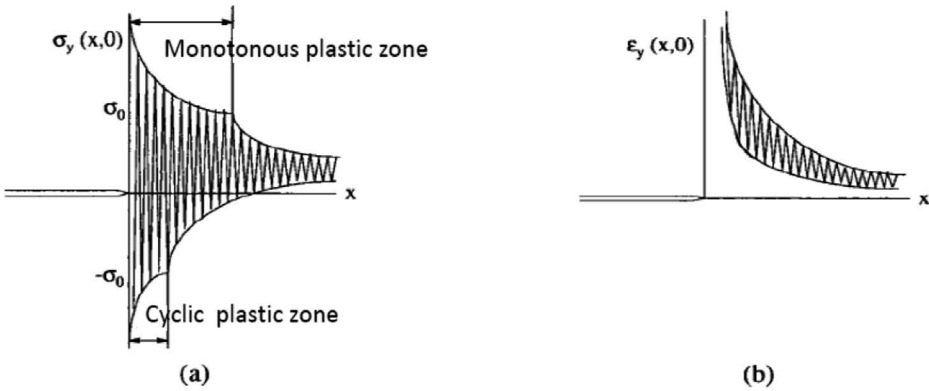


Fig. 1. Evolution for stress vertical component (a) and strain (b) versus crack growth

The cyclic plastic zone size at a crack tip can be written as [35] :

$$r_c = \frac{1}{2\pi\sigma_v^2} (\Delta K)^2 \quad (12)$$

Replacing ΔK obtained from equation (9) and introducing into (12), expression of the radius of the plastic zone is:

$$r_c = \frac{a}{2} \left(\frac{K'(1-a/w)}{\sigma_y} \right)^2 (\Delta \epsilon_p)^{2n'} \quad (13)$$

Substituting equation (13) into equation (8), the crack propagation is given by:

$$\frac{da}{dN} = \frac{a}{2} \left(\frac{K'(1-a/w)}{\sigma_y} \right)^2 (\Delta \epsilon_p)^{(2n'+1)} \quad (14)$$

Introducing equation (4) into equation (14), the relation can be written in the form similar to the Paris law Eq.(1) multiplied by factor β :

$$\frac{da}{dN} = \beta C (\Delta K)^m \quad (15)$$

where the constants β , C and m are defined as:

$$\beta = \frac{(1-a/w)^2}{a^{n'/(1+n')}} \quad (16)$$

$$C = \frac{(2^{n'} K')^2 \sigma_y^{(2n'^2-3n'-3)(1+n')}}{((4\pi + 4\pi n')^{1/(1+n')} E)^{2n'+1}} \quad (17)$$

$$m = \frac{4n'+2}{1+n'} \quad (18)$$

3. APPLICATION AND INTERPRETATION

The theoretical model developed represents the fatigue crack growth based on cyclic hardening parameters (K' , n') and elastics proprieties (E , σ_y) due to the plasticity at the crack tip. Table 1 shows these different parameters found in the literature for two A-alloys studied (2024 T3 and 7075 T6), which were used in the model to obtain the validation and comparison with experimental results [36], [37]. In addition, the variability of the elastics properties and cyclic hardening parameters is evaluated to show their influence on fatigue crack growth.

Table 1. Material and cyclic properties of Al-alloys

Alloys	E [GPa]	σ_y [MPa]	K' [MPa]	n'	Refs
2024 T3	70.3	429	843	0.109	[38]
	73	320	427	0.065	[39]
	73.1	445	576	0.042	[32]
	72	460	590	0.040	[40]
7075 T6	72.2	394	521	0.045	[38]
	71	466	864.1	0.0866	[39]
	71	517	913	0.090	[40]
	70.6	540	790	0.062	[41]
	71	469	977	0,106	[42]
	71.7	468.85	737.81	0.056	[43]
	71	469	781	0.088	[30]

In order to validate the developed model, two applications were given when the compact type C(T) and center-cracked-tension M(T) specimens are used according to ASTM [44] code (E647-00) as shown in Figure 2, with the same applied loading defined in experimental tests [36], [37].

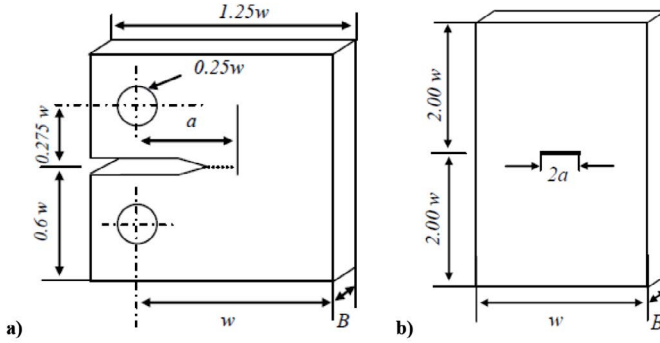


Fig. 2. Schematic of specimens, a) C(T), b) M(T)

The stress intensity factor range ΔK is computed using the applied force as follows:

$$(3.1) \quad \Delta K = \frac{\Delta P}{B} \sqrt{\frac{\pi \alpha}{2w} \sec \frac{\pi \alpha}{2}} \quad \text{with } \alpha = (2a / w) \text{ for specimen M(T) [36]}$$

$$(3.2) \quad \Delta K = \frac{\Delta P}{B} \sqrt{\pi a} f(\alpha') \quad \text{with } \alpha' = (a / w) \text{ for specimen C(T) [35]}$$

$$\text{where, } f(\alpha') = \frac{2 + \alpha'}{(1 - \alpha')^{3/2}} \left[0.886 + 4.64(\alpha') - 13.32(\alpha')^2 + 14.72(\alpha')^3 - 5.6(\alpha')^4 \right]$$

The amplitude loading is given by, $\Delta P = (P_{\max} - P_{\min})$, B and w are respectively the thickness and width of the tested specimens. The geometrical parameters and applied loads for specimens are defined in Table 2.

Table 2. Experimental conditions obtained for tested Al-alloys sample

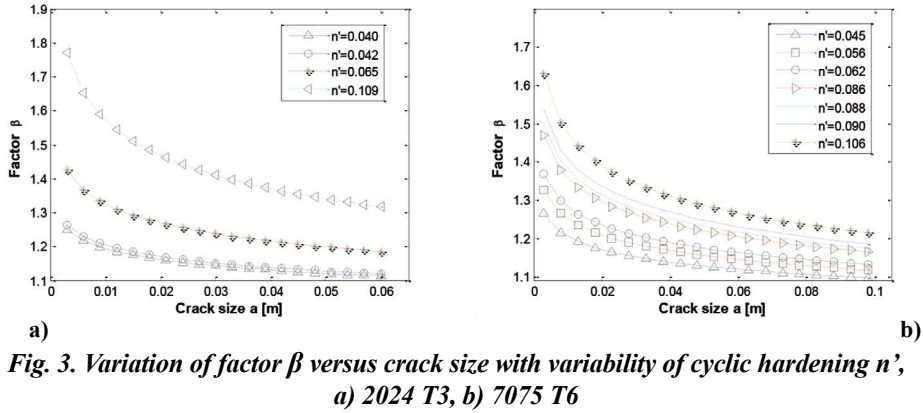
Al-alloy	B [mm]	w [mm]	P_{\max} [N]	P_{\min} [N]	Specimen	Refs
2024 T3	3.2	60.00	2700.0	270.00	C(T)	[37]
7075 T6	3.18	102.03	1902.0	190.20	M(T)	[36]

Firstly, the theoretical evaluation of the plastic zone size in the vicinity of the crack tip is handicapped by the distorted picture of the stress field in front of the fatigue crack. In these respects, the proposed approach has the advantage of being an “interpolation procedure to variability” between the elastic properties (yield strength σ_y , Young’s modulus E) and cyclic hardening parameters (K', n') within the fatigue crack growth rate, hence avoids the risk associated with the inevitable “extrapolation” nature of many other phenomenological but essentially empirical models. These works correctly modeled suggest a qualitative experimental verification, but the model, in the present form, remains essentially speculative.

In the tests of the model for fatigue crack growth rate (Eq. 15) developed by this work, it is of interest to take into account the crack growth effects caused by the different material properties such as Young’s modulus E , yield strength σ_y and cyclic hardening parameters (K', n'). The values of the materials properties listed in Table 1 are plotted against the experimental curves results [36], [37] for the alloys studied, 7075-T6 and 2024-T3, respectively. Figures 4 to 7 show that the experimental and computed values lie near a line drawn through the origin experimental data [36], [37] suggesting a good agreement between them. These plots confirms that fatigue crack growth in the constant ΔK occurred in the Paris regime at an R -ratio of 0.1 consistent with the results of [36], [37] for Al-alloys studied 7075-T6 and 2024-T3, respectively. However, our approach in this study is very direct and perhaps explains the influence of variability for different material properties in a more straightforward way. This variability of the parameters (K', n', σ_y, E) allowed to predict their influence on fatigue crack growth, will be explained below.

The factor β (Eq. 16) is dependent on the normalized crack size and the cyclic strain hardening exponent (n') this factor will be called crack size constant, so is observed by [27]. Figure 3 gives the variation of the factor β as a function of crack growth for various values of (n'). It is observed that for all values of

cyclic hardening exponent (n'), a decreasing of the factor β along the crack propagation (a). Moreover, seems that when ($n' < 0.050$), the factor β becomes stable approximate along the crack (a) about ($1.3 \geq \beta \geq 1.1$) does effect little the crack propagation rate. So when ($n' > 0.050$), this factor take an important value that the first, about ($1.7 \geq \beta \geq 1.2$) is found the crack growth rates increase when cyclic hardening exponent (n') increases. This behavior is in good agreement with the computed curve for identical conditions for Paris's law shown in the following figures.



3.1. Influence of cyclic hardening coefficient (K')

Figures 4a-b show the variability of the cyclic hardening coefficient (K') on the evolution of the fatigue crack growth versus the stress intensity factor range (ΔK) compared with experimental results [36], [37] for alloys studied 7075-T6 and 2024-T3, respectively. The influence of the cyclic hardening coefficient (K') is well marked on the evolution of the fatigue crack growth. It is noticed that the fatigue crack growth increases when the cyclic hardening coefficient increases. Several studies have been remarked this phenomenon [30], [32], [33], [37], [45].

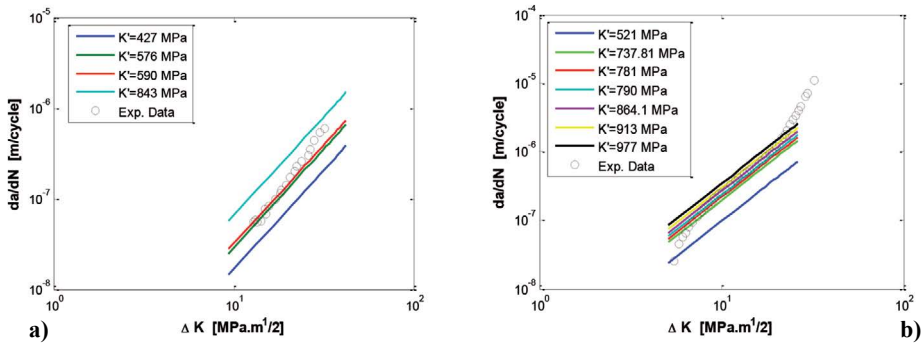


Fig. 4. Fatigue crack growth rate versus stress intensity factor range, variability of cyclic hardening K' , a) 2024 T3, b) 7075-T6

3.2. Influence of cyclic hardening exponent (n')

Other, at the crack tip, in the course of a fatigue cycle, the cyclic plastic zone size effect thermal-mechanical modified material behavior, presents an important hardening. There exist evident influences of the cyclic hardening exponent (n') on the evolution of fatigue crack growth versus stress intensity factor range ΔK . Tow Al-alloys (2024-T3 and 7075-T6) with various cyclic hardening exponent (n') level also analyzed using our proposed model compared with experimental results [36], [37], shown in figure 5a-b, that the fatigue crack growth rates da/dN are a linear function versus the stress intensity factor range ΔK for almost all cyclic hardening exponent (n'). This can be expected if the crack propagates according to the Paris law with respect to the cyclic hardening exponent (n'). The influences to variability clearly of the cyclic hardening exponent (n') increase when fatigue crack growth rate decrease. The hardening effect can be attributed to the influence of plastic strain at some point at the crack front to lead a higher magnitude of plastic strain which governing the evolution of crack growth da/dN versus crack length [30], [32], [33], [37], [45].

For 7075-T6 Al-alloy and at the same stress intensity factor $\Delta K = 13 \text{ MPa}\cdot\text{m}^{1/2}$, the FCGR decreases from $12.9\text{E-}7 \text{ m/cycle}$ to $3.09\text{E-}7 \text{ m/cycle}$ with an increase in the cyclic hardening exponent (n') from 0.045 to 1.106 (about 50% of reduction in cyclic hardening exponent). But for 2024 T3, the FCGR decreases from $1.66\text{E-}7 \text{ m/cycle}$ to $0.42\text{E-}7 \text{ m/cycle}$. In the present study, the developed model (FCGRs model) enclose the experimental using plastic strain energy and the hardening effect contrary to the study conducted by [20], [46]–[48] when the developed models enclose the experimental result with introduce plastic strain energy and the crack closure effect. The variability of the cyclic hardening exponent show a moderate effect on fatigue crack growth rates (FCGR) is significant for 7075 T6 Al-Alloy comparatively to 2024 T3 Al-Alloy.

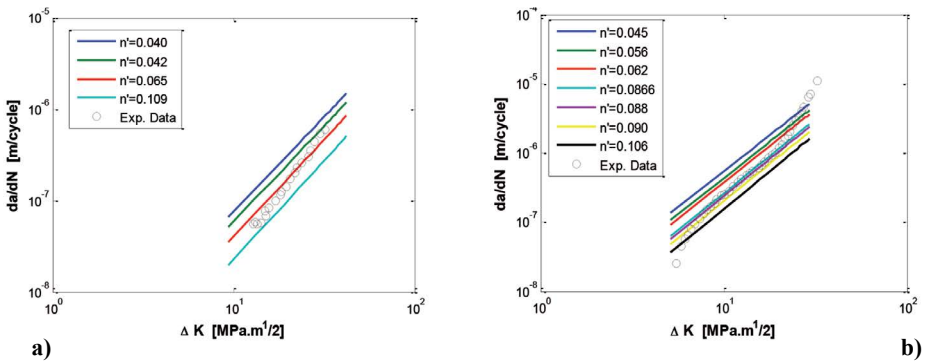


Fig. 5. Fatigue crack growth rate versus stress intensity factor range, variability of cyclic hardening exponent (n'), a) 2024 T3, b) 7075 T6

3.3. Influence of cyclic yield strength (σ_y)

Figure 6a-b show the fatigue crack growth rate versus the stress intensity factor range with different values of yield strength into relationship (Eq.15). The influence to variability it seems clearly suitable thus the velocity of fatigue crack growth increase when the yield strength decreases [17], [32], [45], [47], [49]. However, we can say that the yield strength is an important parameter into this model for the fatigue crack growth rate (Eq.15). It confirms that the yield strength has stronger influence on the fatigue crack growth.

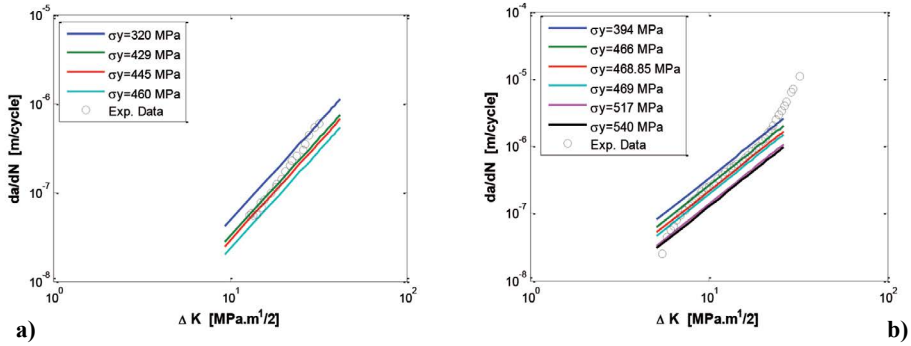


Fig. 6. Fatigue crack growth rate versus stress intensity factor range, variability of yield strength σ_y , a) 2024 T3, b) 7075 T6

3.4. Influence of Young's modulus (E)

The fatigue crack growth rates for Aluminum alloys are much more rapid than those in steel for a given ΔK [35], [36], [45]–[47]. Figures 7a-b show the fatigue crack growth rate as a function the stress intensity range ΔK at different values of Young's modulus compared with the experimental results [36], [37] for the studied Al-alloys, 7075-T6 and 2024-T3, respectively.

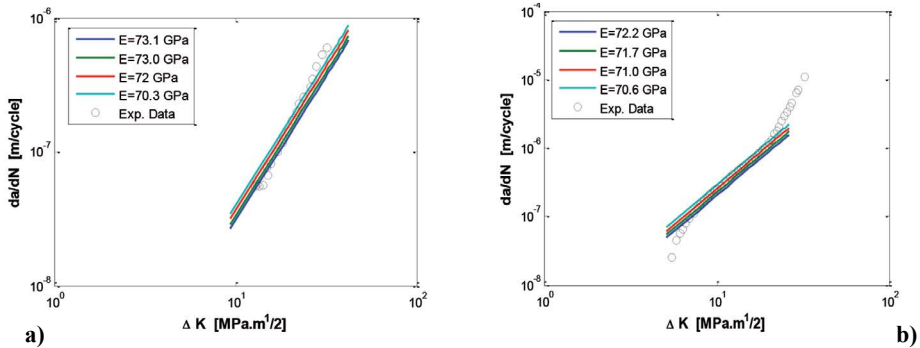


Fig. 7. Fatigue crack growth rate versus stress intensity factor range, variability of cyclic Young's modulus E, a) 2024 T3, b) 7075 T6

It is shown that there exist weakly influence by variability to this parameter such as the fatigue crack growth rate increases slowly when the Young's modulus decreases [45], [47], [49]. Others, observed the fatigue threshold models predict a strong influence of the elastic modulus [50], [51].

According to the differentiation between two types of behavior of Al-alloys introduced above table 2, the comparison of values of the material constants C and m between the obtained from this model (Eq.17 and 18) for different material properties studied (K' , n' , σ_y , E) with the average values of literature realized a comparison acceptable for same R -ratio $R=0.1$, as shown table 3.

Table 3. Modeling and experimental values of parameters of Paris law

Alloys	Modeling values				Experimental values			
	n'	C [m/cycle]	m	R	C [m/cycle]	m	R	Refs
2024 T3	0.040	6.2815E-10	2.0769	0.1	7.5100E-10	2.01	0.1	[37]
	0.042	5.9242E-10	2.0806					
	0.065	3.0259E-10	2.1221		2.6029E-10	2.65	0.1	[52]
	0.109	8.4245E-11	2.1966					
7075 T6	0.045	4.1364E-09	2.0861	0.1	1.5739E-09	2.19	0.1	[36]
	0.056	3.0011E-09	2.1061					
	0.062	2.5199E-09	2.1168					
	0.086	1.2328E-09	2.1594		1.6500E-10	3.00	0.1	[4]
	0.088	1.1837E-09	2.1618					
	0.090	1.1170E-09	2.1651					
	0.106	7.0248E-10	2.1917					

Others, explained a decrease in the value of the exponent (m) of the Paris law when the yield strength (σ_y) increased [35], [53], [54]. So the exponent (m) changes slightly at different temperatures, while constant (C) for the Paris law changes with the temperature [55]. The analysis of these results shown in Table 3 also reveals an increase in the values of (m) when the cyclic hardening exponent (n') increases, and strictly inverted of constant (C) of Paris law. Figure 8 shows the evolution of the constant (C) for the Paris law versus the cyclic hardening coefficient (K'), we have remarked the variability of the coefficient (K') increase, when the constant (C) increases.

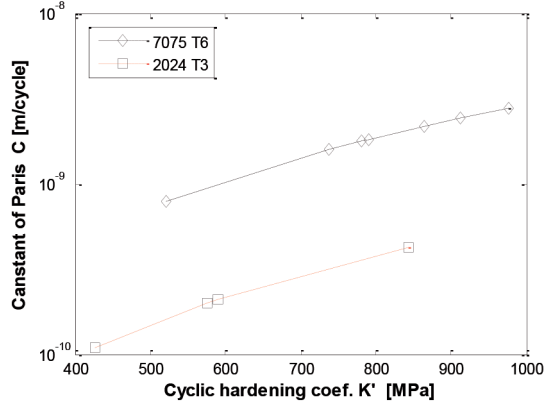


Fig 9. Constant of Paris law C versus cyclic hardening coefficient K'

Using the relationship (Eq.17), the influence of elastic properties (Yield strength σ_y and Young's modulus E) on the constant C of the Paris law is shown in Fig. 9. The constant C is plotted against the yield strength in (Fig. 9-a) explain that when the variability of yield strength (σ_y) increase, when the constant C decrease.

The Paris exponent (m) decreases when increasing the Young's modulus, is remarked by [51], the (Fig 10-b) shows the evaluation of the constant C versus the Young's modulus, seems linear, as approximate stable, the variability of Young's modulus is influenced little the constant C .

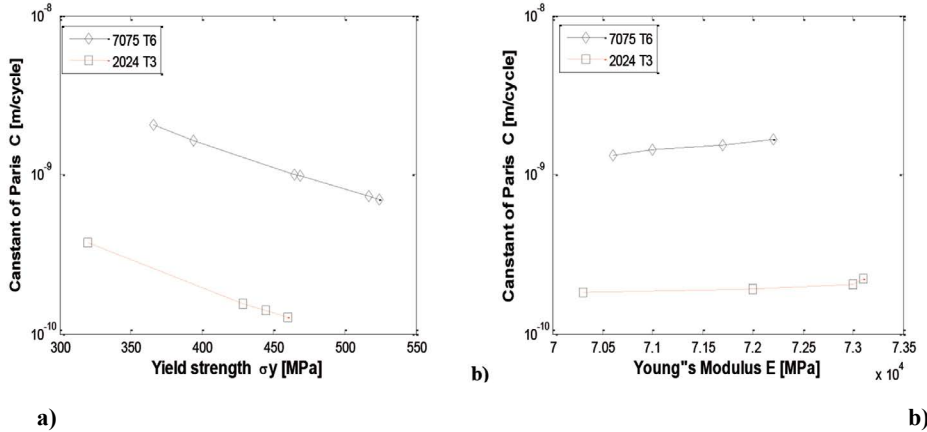


Fig. 10. Constant C of Paris law versus, a) yield strength, b) Young's modulus

4. CONCLUSION

This paper focuses a contribution in development of analytical model for fatigue crack growth in Paris stage as a function of different materials properties (Yield strength, Young's modulus and cyclic hardening parameters) based on the cyclic plastic deformation at the crack tip. The experimental data are presented in this paper for the validation of the developed model.

The main advantage of the developed model over the literature model is the direct dependence of the Paris law constants (C and m) on mechanical properties and hardening parameters. Also, a separate parameter is included in the Paris law characterized by the constant (b) depending only on the hardening parameter (n') and the ratio (a/w) used in the correction function of geometry specimen (see equations 15 and 16). The developed model is applied to stable stage (Paris region) comparatively to other models such as Duggan, Forman and Tzamtzis models, which include the instable stage of crack growth.

Thus, from this study, the mechanical properties and hardening parameters effects are investigated on fatigue crack growth rate and the following conclusions can be drawn:

- The experimental data use for validation is enclosed by the developed model with good agreement.
- The fatigue crack growth rate increase in an increasing the hardening parameter (K') and their effect is shown clearly in the evolution of constant (C) in the Paris law.
- The variability of cyclic hardening exponent (n') shows a moderate effect on FCGRs where the effect of hardening exponent (n') is significant on FCGRs of 7075 T6 Al-Alloy comparatively to 2024 T3 Al-Alloy.
- Fatigue crack growth rate increases with decreasing the yield strength.
- Slow increasing of FCGRs is noticed with a decreasing in Young's modulus.
- The constants (C , m) of the Paris law depend strongly on the cyclic hardening exponent.

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